

A B S T R A C T

A DAMPER FOR A LANDING GEAR LEG, AND LANDING GEAR HAVING
INDEPENDENT LEGS FITTED WITH SUCH DAMPERS

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The invention relates to a damper for an airplane
landing gear leg, the damper being of the type comprising
a main strut (11) and a rod-piston (13) co-operating with
the strut (11) to define a main chamber (15) and annular
10 chamber (16) for hydraulic fluid, and presenting
internally two adjacent chambers (19, 20) that are
isolated from each other by a separator piston (21). In
accordance with the invention, the damper (10) further
comprises a first secondary strut (26) telescopically
15 slidable on the above-mentioned rod-piston (13), and a
second secondary strut (37) telescopically slidable on
the other end of the first secondary strut (26). The two
second annular chambers (31, 40) as defined in this way
are respectively connected to associated control circuits
20 thus enabling the total length of the damper to be
shortened or lengthened respectively for the purpose of
causing the landing gear leg to contract or to be
extended.

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Translation of the title and the abstract as they were when originally filed by the
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